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Galorath –A Technical Solutions Provider

AN ESOP Company Dedicated to quality (ISO 9001:2008 certified)





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- Overview of Galorath Inc.
- Overview of SEER Models
- SEER-Space
- Demo of SEER-Space

Galorath Background



- Galorath Incorporated has invested more than three decades developing solutions to help government and commercial organizations plan and manage complex projects.
- Introduced SEER commercial software in 1988
- Activities include:
 - SEER parametric tool development and support
 - SEER implementation services
 - SEER product training
 - Professional cost analysis / management consulting services
 - Custom parametric tool development
- SEER solutions combine an intuitive interface, extensive project-applicable Knowledge Bases, sophisticated project-modeling technologies, and rich reporting features to expedite the planning process and keep projects on track.
- Worldwide clients in:
 - Defense / aerospace
 - Manufacturing
 - Finance, insurance, consulting

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An Overview of SEER Models



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SEER Suite of Applications



• Different products for different domains/disciplines



SEER-SEM – software/application development, maintenance, integration and testing



SEER-H – system, hardware and electronics development, production and support



SEER-IT – IT infrastructure, services and operations



SEER-MFG – hardware manufacturing and assembly



SEER-SYS – Systems Engineering



SEER-Space – New!

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SEER-Space



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Why SEER-Space?



- SEER-Space (Version 1) is NASA centric in terms of the data behind the model
 - However, it is applicable to non-NASA space missions as well
 - Space projects all use the same Physics
 - Same Aerospace Engineering processes
 - Same systems, subsystems and components
 - Similar institutions and contractors
- And SEER-Space includes cost drivers which are known to be good predictors of space project costs, both quantitative variables and some that are qualitative
 - Balances the art and science of cost modeling
- SEER-Space can be used in very early conceptual stages of the project life cycle
 - But is also applicable through mid life cycle (PDR to CDR)
 - Early utility is further supported by SEER-Space Knowledge Bases

Differentiation of SEER-Space From SEER-H/SEM SEM



| | SEER-H and SEM | SEER-Space |
|------------------------------------|--|--|
| Level of detail | MEL level (or close) | Bus subsystems + instruments |
| Work Breakdown Structure | NASA WBS Elements 1 through 6 + 10 (NR + Production) | NASA WBS Elements 1-11 (NR + Production + Operations) and DOD MIL- STD-881D |
| Applicability to Space Missions | More generic Kbases and input parameters that are applicable to both Space and Non-Space Projects | NASA Class A, B, C and D Standards, Technology Readiness Levels effects, Earth Orbital and Deep Space missions (including Orbiters, Flybys, Landers, and Rovers) |
| Database | Various data sources + SME derived CERs | Recent missions drawn from the NASA CADRe database as well as additional data sources and extensive research by Galorath analysts |
| CER derivations | Various including SME derived | Regression analysis including cross-validation |

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- SEER-Space will estimate all 11 NASA WBS elements <u>plus</u> Phase E Mission Operations and Data Analysis (MO&DA) [aka Phase E operations cost] and DOD MIL-STD-881D
- The model allows for the accounting of hardware contributions made to a mission (from part level to whole instruments)
- Supports the analysis of different means of acquisition such as competed versus directed projects, prime contractor versus university-built, foreign contributions, and more.
- It can also assess heritage benefits (major modification vs minor modification) and Technology Readiness Level (TRL)





- SEER-Space knowledge bases allow you to create rough order of magnitude estimates with only a few high level inputs.
- SEER-Space offers five basic categories of knowledge bases
 - Applications
 - Project Rollup
 - Bus
 - Instrument
 - Telescope
 - Data Processing Unit
 - Cryocooler
 - Platform
 - Heritage
 - Standards
 - Organization

| Bus: Structures and Mechanisms | Least | Likely | Most | |
|---|----------|-------------------|----------|--|
| STRUCTURAL/MECHANICAL SUBSYSTEM | | Yes | | |
| Mass (kg) (SMS) | 1,000.00 | 1,000.00 | 1,000.00 | |
| Number Of Deployables (SMS) | 2 | 3 | 5 | |
| Deployable Complexity (SMS) | Hi+ | VHi- | VHi | |
| New Development (SMS) | 50.00% | 60.00% | 70.00% | |
| Technology Complexity (SMS) | Nom- | Nom | Nom+ | |
| - THERMAL CONTROL SUBSYSTEM (TCS) | | Yes | | |
| Mass (kg) (TCS) | 25.00 | 25.00 | 25.00 | |
| Thermal Control Type (TCS) | | Active | | |
| New Development (TCS) | 50.00% | 60.00% | 70.00% | |
| Technology Complexity (TCS) | Nom- | Nom | Nom+ | |
| - COMMUNICATIONS, COMMAND & DATA H | | Yes | | |
| Mass (kg) (CCDH) | 50.00 | 50.00 | 50.00 | |
| Number of Bands (CCDH) | 2 | 3 | 4 | |
| | | Yes | | |
| Redundancy (CCDH) | | Partial | | |
| New Development (CCDH) | 50.00% | 60.00% | 70.00% | |
| Technology Complexity (CCDH) | Nom- | Nom | Nom+ | |
| - GUIDANCE, NAVIGATION & CONTROL SU | | Yes | | |
| Mass (kg) (GNC) | 100.00 | 100.00 | 100.00 | |
| Control Mechanism (GNC) | | 3-Axis Controlled | | |
| Redundancy (GNC) | | Partial | | |
| | | No | | |
| New Development (GNC) | 50.00% | 60.00% | 70.00% | |
| Technology Complexity (GNC) | Nom- | Nom | Nom+ | |
| - ELECTRICAL POWER SUBSYSTEM (EPS) | | Yes | | |
| Mass (kg) (EPS) | 75.00 | 75.00 | 75.00 | |
| Solar Array Type (EPS) | | Deployable | | |
| Power (W) (EPS) | 0.00 | 0.00 | 0.00 | |
| Total Radioisotopic Thermal Generators (E | 0 | 0 | 0 | |
| New Development (EPS) | 50.00% | 60.00% | 70.00% | |
| Technology Complexity (EPS) | Nom- | Nom | Nom+ | |
| -REACTION CONTROL SUBSYSTEM (RCS) | | Yes | | |
| Mass (kg) (RCS) | 0.00 | 0.00 | 0.00 | |
| Propulsion Type (RCS) | | Mono-Propellant | | |
| New Development (RCS) | 50.00% | 60.00% | 70.00% | |
| Technology Complexity (RCS) | Nom- | Nom | Nom+ | |

Instrument Application Kbases



- ~Fields General
- ~Optical General
- ~Particle General
- ~Radio General
- ~Sample Acquisition General
- ~Sample Analysis General
- Calorimeter
- Camera Multispectral Imager
- Camera Spectral Imager
- Chemical Microscopy Scanner
- Gas Chromatograph
- Electric Field Sensor
- Compound Instrument- Multispectral Imager
- Langmuir Probe
- LASER Altimeter
- LASER LIDAR
- LASER Optical Transceiver
- LASER Spectrometer
- Magnetometer Fluxgate
- Magnetometer Search Coil
- Particle Ion Trap Mass Spectrometer
- Particle Quadrupole Mass Analyzer © 2018 Copyright Galorath Incorporated

- Particle Sector Mass Spectrometer
- Particle Time of Flight Mass Spectrometer
- Particle Detector, Dust
- Particle Detector, Energetic Particle
- Particle Detector, Ion/Plasma
- RADAR Scatterometer
- RADAR Synthetic Aperture
- RADAR Altimeter
- Radio Antenna
- Radio Receiver
- Radio Transceiver
- Radiometer IR
- Radiometer Microwave
- Radiometer UV/Visible
- Spectrograph IR
- Spectrograph UV/Visible
- Spectrograph X Ray/Gamma Ray
- Spectrometer IR
- Spectrometer UV/Visible
- Spectrometer X Ray/Gamma Ray
- Spectroradiometer

Data Collection



- Data are the foundation of cost estimates
- Researchers at Google found that simple models based on a lot of data are better than more elaborate models based on less data
- Need sound, quantitative data
 - Cost
 - Technical
 - Programmatic

Analysts spend the majority of their time developing techniques and honing tools, when the most important focus should be on the quality and quantity of data.



Data Normalization



- For some older missions adjusted to account for fullcost accounting, as applicable
- Split all costs into nonrecurring and recurring costs
- Normalized cost to a constant base year using latest NASA inflation guidance
- For multiple units, normalized recurring costs to a theoretical first unit cost

Data and CERs Included



- SEER-Space includes new cost estimating relationships(CERs) for bus subsystems and instruments, including dedicated CERs for optical telescopes and data processing units
- Collected and normalized data from recent CADRe files
- Assumed that data for earth-orbiting and planetary missions can be used together for CER development

 can include a dummy variable (aka binary/indicator variable) to account for the difference in the model
- Bus subsystem CERs based on ~50 recent earthorbiting and planetary missions
- Instrument CERs based on ~150 data points from recent earth-orbiting and planetary missions

CER Development Methodology



- The oldest method still in common use for developing power-form CERs is log-transformed ordinary least squares linear regression (LOLS)
- However this method has been criticized for many years for multiple reasons:
 - Transformation causes equation to be optimal for "logdollars"
 - Result is biased (estimating the median vice the mean)
- Other methods such as Minimum Unbiased Percentage Error (MUPE) and Zero-bias Minimum Percent Error (ZMPE) have been recommended as alternatives to logtransformed linear regression
- However, statistical analysis of residuals for spacecraft CERs has shown that the best for the residuals is the lognormal distribution

CER Development Methodology (2)

- Developed a new CER method that has the advantages of log-transformed linear regression without the transformation issues
- Used solver to minimize a maximum likelihood equation that is optimal for lognormal residuals
- Similar to LOLS but unbiased
- No transformation required
- Also set logical constraints on the values for the coefficients

Model Form Application



- For SEER for Space, we used the power equation for most subsystems and instruments
- In our experience power forms have worked well for other models, and they fitted the data well for this project so we used them extensively
- All CERs are multivariate
- In some cases when the cost of the system was low, we used linear equations (e.g., data processing units)
 - When the cost is low, it has a limited range as the lower bound is equal to zero
 - Log transformations compress the spread of the data, so low cost items are typically not nonlinear

Model Inputs



- All CERs include a set of programmatic parameters:
 - AO or Directed
 - Earth Orbiting or Planetary
 - More than one sponsor
 - International Involvement
 - Extensive Testing
 - Number of Instruments
 - Number of Active Instruments
 - Design Life (Months)
- The number of instruments and number of active instruments is not applied to the instrument CERs
- All instrument CERs except for the telescope CER include power as a driver

Model Inputs (2)



- In addition to the common factors, there are several CERs that include unique technical independent variables
 - Telescope Mirror diameter is the primary driver, weight is not an independent variable
 - Bus subsystems:
 - Structures: number of deployables and deployable complexity
 - Thermal: type (passive/active)
 - CCDH: number of bands, high-gain antenna (yes/no), and level of redundancy
 - GNC: 3-axis (yes/no), level of redundancy, and start tracker (yes/no)
 - Power: deployable array (yes/no), maximum power
 - Reaction control: monopropellant (yes/no)

Variable Selection



- The primary CER variable that we use is weight (aka mass)
- This is not a causative driver of cost, merely a scaling parameter
- Our models are not based on "cost drivers"
- However these parameters explains historical variation well and we typically have estimates of weight early in a program's lifecycle so we continue to use these parameters

Variable Selection (2)



- Another controversial topic is the use of subjective parameters such as heritage (or its reciprocal, percent new design)
 - The amount of new design is strongly correlated with the nonrecurring cost of a program
 - Heritage is typically overrated early in a program's lifecycle, can lead to underestimation
 - The subjective nature can lead to misestimation when compared to the historical cost
 - However, there is a need by customers to discern the impact of new design on cost, so we have made the decision to include new design as a parameter in SEER for Space

Cross Validation



All CERs were developed using cross-validation

- In-sample standard errors will necessarily be lower than the generalization standard error
- We can measure this as we develop the model by conducting n-fold cross-validation
- N-fold cross validation involves splitting the data set into multiple partitions (n-folds)
- Train the model on n-1 of the partitions and test on the remainder
- Repeat this process n times, once for each fold
- Average the results
- N-fold cross validation allows the validation error bounds to be calculated as a result of the CER development process
- N-fold cross validation has the advantage in small data sets of saving more of the data for training



Space Systems Cost Estimation Workshop



- Space domain-focused workshop provides knowledge and skills necessary to estimate the cost, schedule and effort of space systems
- Students given overview of different methods of estimation being used today: including grass-roots and bottom-up estimates, analogy-based estimates, system level parametric models
- Led through a discussion of the differences between system level parametric models, such as the NASA Instrument Cost Model (NICM), and component level parametric models, such as SEER-H.
- Learn differences between modeling various spacecraft subsystems, how to model various instrument types and gain insights as to how to estimate new technology developments
- Familiarized with assessing cost, schedule, risk and how to build a defendable estimate
- Upon completion, will be able to produce a cost estimate for the full life cycle of a space mission project, from pre-concept through operations, taking into account the risks, uncertainty and cost drivers associated with space hardware development, acquisition and integration.



- Instructors don't just teach— extensively involved with SEER-H, NICM, PCEC, MOCET, Quickcost, SEER for Space Systems and other Space cost models.
- Instructors developed CERs with NASA and DoD and will augment training with "lessons learned" from their experience





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